

IN THE SPECIFICATION

Please delete paragraph [0002] and replace the following therefor:

Gas turbine engine rotor blades typically include airfoils having leading and trailing edges, a pressure side, and a suction side. The pressure and suction sides are connected at the airfoil leading and trailing edges, and span radially between the airfoil root and the tip. During operation, combustion gases impacting impact the rotating rotor blades transfer transferring heat into the airfoils. Over time, continued exposure to high temperature combustion gases may thermally fatigue the airfoil.

Please delete paragraph [0003] and replace the following therefor:

To facilitate preventing damage ~~due~~ to the airfoils from exposure to the high temperature combustion gases, known airfoils include an internal cooling circuit which channels cooling fluid through the airfoil. Specifically, at least some known rotor blades channel compressor bleed air into a cavity defined between the sidewalls, to convectively cool the sidewalls. Moreover, at least some known cooling circuits utilize shear-jet cooling wherein a plurality of shear-jet openings channel cooling fluid along an inner surface of the sidewalls to facilitate cooling the sidewalls. Additional cooling cavities can be accomplished using impingement cooling wherein impingement inserts channel cooling fluid through impingement jet arrays against the inner surface of the airfoil's leading edge to facilitate cooling the airfoil along the leading edge. However, these circuits are inefficient as the circuits allow the cooling fluid to flow through the center of the cavity where it is ineffective in removing heat from the walls of the airfoil.